

Amendments to the Claims

Claims 1-5. (Cancelled)

Claim 6. (New) A rotor for a gas turbine, having blades arranged on the rotor and rotating together with the rotor, the blades forming a blade ring and the blades within the blade ring being arranged at a different distance from one another and thus with a different blade pitch, a distance between the blades within the blade ring changing continuously or discontinuously in a circumferential direction and the distance between the blades within the blade ring dimensioned such that an imbalance is canceled out, wherein the rotor has several blade rings arranged axially behind one another, and wherein, within each blade ring, the blades are arranged at a different distance from one another.

Claim 7. (New) The rotor according to Claim 6, wherein an arrangement of blades within a respective blade ring is different for each of the blade rings.

Claim 8. (New) The rotor according to Claim 6, wherein the rotor is a turbine rotor or compressor rotor of a gas turbine.

Claim 9. (New) The rotor according to Claim 8, wherein the gas turbine is a turbine of an airplane engine.

Claim 10. (New) The rotor according to Claim 6, wherein the rotor is a fan rotor of a gas turbine.

Claim 11. (New) The rotor according to Claim 10, wherein the gas turbine is a turbine of an airplane engine.

Claim 12. (New) The rotor according to Claim 6, wherein the rotor is a blisk (bladed disk) or bling (bladed ring) of a gas turbine and wherein the blades are an integral component of the rotor.

Claim 13. (New) The rotor according to Claim 12, wherein the gas turbine is a turbine of an airplane engine.

Claim 14. (New) A rotor for a gas turbine, comprising:
a first blade ring; and
a second blade ring;
wherein the first and the second blade rings are arranged axially one behind the other and wherein on each blade ring a first blade is positioned at a first distance from a second blade and the second blade is positioned at a second distance from a third blade.

Claim 15. (New) The rotor according to Claim 14, wherein on each blade ring a fourth blade is positioned at the first distance from a fifth blade and the fifth blade is positioned at the second distance from a sixth blade and wherein the first, second, and third blades are located diametrically opposed from the fourth, fifth, and sixth blades, respectively, on the blade ring.

Claim 16. (New) The rotor according to Claim 14, wherein an arrangement of blades within a respective blade ring is different for each of the blade rings.

Claim 17. (New) A method for optimizing vibrations in a gas turbine engine, comprising the steps of:
rotating a first blade ring of a rotor;
rotating a second blade ring of a rotor;
wherein on each blade ring a first blade is positioned at a first distance from a second blade and the second blade is positioned at a second distance from a third blade; and
continuously changing a frequency of a vibration of a stationary assembly of the gas turbine by rotating the first and second blade rings.